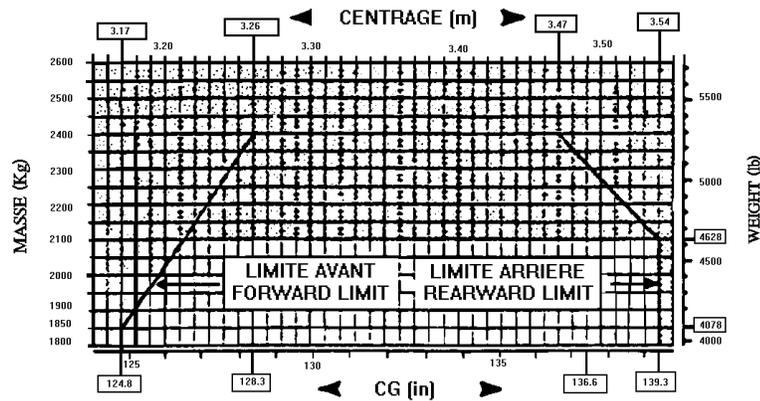


C.G. Range

Longitudinal

Datum.

Longitudinal = 133.8 in. forward of main rotor hub center

Lateral = Vertical plane passing longitudinally through main rotor hub center

Maximum Weight.

4628 lb (2100 kg)

Rotor Speed.

Power-off

maximum = 425 rpm

minimum = 330 rpm

Power-on

= 390 rpm with both engine operating

= 375 to 394 rpm with one engine inoperative

Rotor Speed Warning.

Aural at 360 rpm

Airspeed Limits.Never exceed speed V_{NE} Power-on:

0 - 73% torque. 150 kts at zero pressure altitude.

Doors removed, 70 kts at zero pressure altitude.

73% - 78% torque. 55 kts at all altitudes.

Never exceed speed V_{NE} Power-off:

120 kt at zero pressure-altitude

See Rotorcraft Flight Manual for decrease of these values with altitude and temperature.

Minimum Crew.

1 pilot at 60.62 in. - Starboard seat

Maximum Passenger. (5).

4 at 98.42 in.

1 at 60.62 in.

Maximum Baggage.

Right baggage compartment = 220 lb at 125.98 in

Left baggage compartment = 264 lb at 125.908 in

Rear baggage compartment = 176 lb at 181.10 in

Main cabin (on rear = 682 lb at 88.58 in

(on LH forward = 330 lb at 61.02 in

Fuel Capacity.

	Front Tank	Rear Tank
Total	88.1 U.S. gallons	106.5 U.S. gallons
Usable	87.2 U.S. gallons	105.7 U.S. gallons
C.G. Range	127.55 in	151.55 in

See NOTE 1 for data on unusable fuel.

Empty Weight CG Range.

None

Leveling Means.

Transmission support platform.

Oil Capacity.

Engine = 2 x 1.5 U.S. gallons at 140.5 in
 MGB = 2.9 U.S. gallons at 145.2 in
 TGB = 0.08 U.S. gallons at 379.5 in

Rotor Blades and Control Movements.

For rigging information, refer to the AS 355 Maintenance Manual.

II. Model AS 355-F "TWINSTAR" (Normal Category), November 20, 1981.

(Same as model AS355E except for maximum weight and dual hydraulic systems).

Engine.

2 ALLISON Model 250-C20F.

Installed Engine Limits.

	Torque	Gas Generator Speed	Exhaust Gas Temp.	Output Shaft Speed
Take-off (5 min) (two engines operative)	380 mN (73%) 380 mN (73%)	53519 rpm "	810°C 738°C	6196 rpm "
Maximum Continuous (one engine operative)	521 mN (100%)	"	810°C	"

Engine gearbox torque limit = 384 ft=lb (521 mN) = 100%

Refer to Flight Manual AS355F for transients.

Fuel.

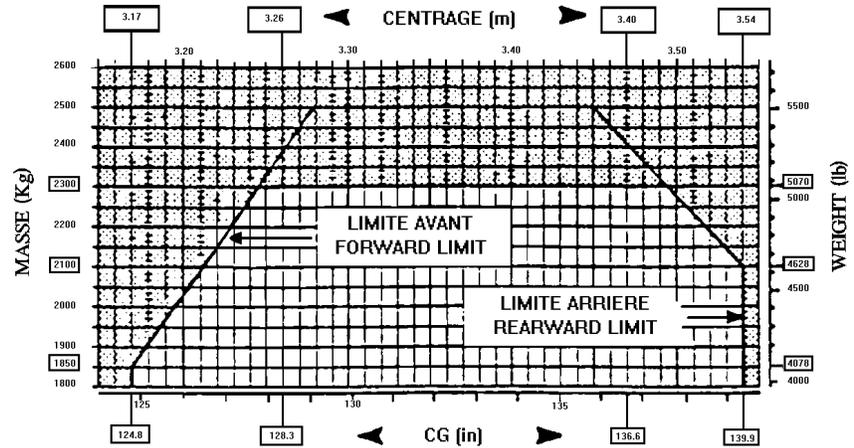
Refer to Flight Manual AS355F for approved fuels and additive specification.

Helicopter Limits.

Maximum (one engine inoperative): 100% (521 mN)
 Maximum (two engines operative): 73% (380 mN)

C.G. Range.

Longitudinal



Lateral - Right = 3.54 in
 Left = 6.30 in

Datum.

Longitudinal = 133.8 in. forward of main rotor hub center
 Lateral = Vertical plane passing longitudinally through main rotor hub center

Maximum Weight.

5070 lb (2300 kg)

Rotor Speed.

Power-off
 maximum = 425 rpm
 minimum = 330 rpm

Power-on
 = 390 rpm with both engines operating
 = 375 to 394 rpm with one engine inoperative

Rotor Speed Warning.

Aural at 360 rpm

Airspeed Limits.

Never exceed speed V_{NE} Power-on:
 150 kt at zero pressure altitude. Doors removed; 70 knots at zero pressure altitude.

Never exceed speed V_{NE} Power-off

120 kt at zero pressure-altitude
 See Rotorcraft Flight Manual for decrease of these values with altitude and temperature.

Minimum Crew.

1 pilot at 60.62 in. - Starboard seat

Maximum Passengers (5).

4 at 98.42 in.
 1 at 60.62 in.

Maximum Baggage.

Right baggage compartment = 220 lb at 125.98 in
 Left baggage compartment = 264 lb at 125.908 in
 Rear baggage compartment = 176 lb at 181.10 in
 Main cabin (on rear = 682 lb at 88.58 in
 (on LH forward = 330 lb at 61.02 in)

Fuel Capacity.

	Front Tank	Rear Tank
Total	88.1 U.S. gallons	106.5 U.S. gallons
Usable	87.2 U.S. gallons	105.7 U.S. gallons
C.G. Range	127.55 in	151.55 in

See NOTE 1 for data on unusable fuel.

Empty Weight CG Range.

None

Leveling Means.

Transmission support platform.

Oil Capacity.

Engine = 2 x 1.5 U.S. gallons at 140.5 in
 MGB = 2.9 U.S. gallons at 145.2 in
 TGB = 0.08 U.S. gallons at 379.5 in

Rotor Blades and Control Movements.

For rigging information, refer to the AS 355 Maintenance Manual.

III. Model AS 355-F1 "TWINSTAR" (Normal Category), April 11, 1984.Engine.

2 ALLISON Model 250-C20F

Installed Engine Limits.

	Torque	Gas Generator Speed	Exhaust Gas Temp.	Output Shaft Speed
Take-off (5 min)	406 mN (78%)	53,519 rpm	810°C	6,196 rpm
Maximum Continuous (two engines operative)	380 mN (73%)	"	738°C	"
(one engine operative)	521 mN (100%)	"	810°C	"

Engine gearbox Torque limit = 384 ft=lb (521 mN) = 100%

Refer to Flight Manual AS355F1 for transients.

Fuel.

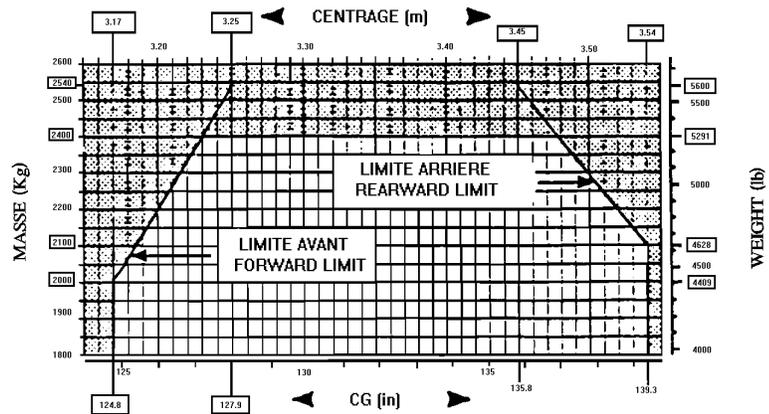
Refer to Flight Manual AS355F1 for approved fuels and additive specifications.

Helicopter Limits.

Maximum (one engine inoperative): 100% (521 mN)
 Maximum (two engines operative): 78% (406 mN)

C.G. Range.

Longitudinal



Lateral - Right = 3.54 in
 - Left = 6.30 in

Datum.

Longitudinal = 133.8 in. forward of main rotor hub center
 Lateral = Vertical plane passing longitudinally through main rotor hub center.

Maximum Weight.

5291 lb (2400 kg)

Rotor Speed.

Power-off
 maximum = 425 rpm
 minimum = 330 rpm
 Power-on
 = 390 rpm with both engines operating
 = 375 to 394 rpm with one engine inoperative

Rotor Speed Warning.

Aural at 360 rpm and 410 rpm.

Airspeed Limits.

Never exceed speed V_{NE} Power-on:
 150 kt at zero pressure altitude. Doors removed; 70 knots at zero pressure altitude.
 Never exceed speed V_{NE} Power-off
 120 kt at zero pressure-altitude
 See Rotorcraft Flight Manual for decrease of these values with altitude and temperature.

Minimum Crew.

1 pilot at 60.62 in. - Starboard seat

Maximum Passengers Standard(5).

4 at 98.42 in.
 1 at 60.62 in.
 Maximum passengers of 6 is available when the aircraft is equipped with the optional two-place seat at 60.62 in. This optional item is to be used in accordance with the associated Flight Manual supplement.

Maximum Baggage.

Right baggage compartment = 220 lb at 125.98 in
 Left baggage compartment = 264 lb at 125.908 in
 Rear baggage compartment = 176 lb at 181.10 in
 Main cabin (on rear = 682 lb at 88.58 in
 (on LH forward = 330 lb at 61.02 in

Fuel Capacity.

	Front Tank	Rear Tank
Total	88.1 U.S. gallons	106.5 U.S. gallons
Usable	87.2 U.S. gallons	105.7 U.S. gallons
C.G. Range	127.55 in	151.55 in

See NOTE for data on unusable fuel.

Empty Weight CG Range.

None

Leveling Means.

Transmission support platform

Oil Capacity.

Engine = 2 x 1.5 U.S. gallons at 140.5 in
 MGB = 2.9 U.S. gallons at 145.2 in
 TGB = 0.08 U.S. gallons at 379.5 in

Rotor Blades and Control Movements.

For rigging information, refer to the AS 355 Maintenance Manual.

IV. Model AS355F2 "TWINSTAR" (Normal Category), February 13, 1987.

(Same as AS355F1 except for maximum weight, external load maximum weight, yaw compensation system, etc.)

Engine.

2 ALLISON Model 250-C20F

Installed Engine Limits.

	Torque	Gas Generator Speed	Exhaust Gas Temp.	Output Shaft Speed
Take-off (5 min)	406 mN (78%)	53,519 rpm	810°C	6,196 rpm
Maximum Continuous (two engines operative)	380 mN (73%)	"	738°C	"
(one engine operative)	521 mN (100%)	"	810°C	"

Engine gearbox Torque limit = 384 ft=lb (521 mN) = 100%

Refer to Flight Manual AS355F2 for transients.

Fuel.

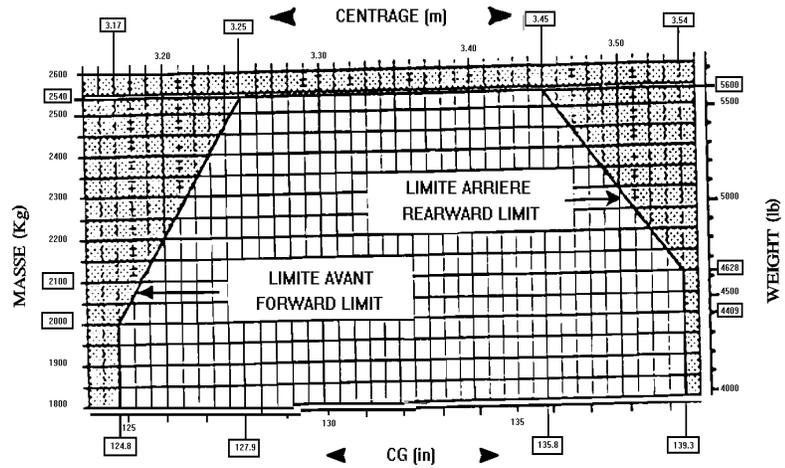
Refer to Flight Manual AS355F2 approved fuels and additive specification.

Helicopter Limits.

Maximum (one engine inoperative): 100% (521 mN)
 Maximum (two engines operative): 78% (406 mN)

C.G. Range.

Longitudinal



Lateral - Right = 3.54 in
 - Left = 6.30 in

Datum.

Longitudinal = 133.8 in. forward of main rotor hub center
 Lateral = Vertical plane passing longitudinally through main rotor hub center

Maximum Weight.

5600 lb (2540 Kg)

Rotor Speed.

Power-off
 maximum = 425 rpm
 minimum = 330 rpm

Power-on
 = 390 rpm with both engines operating
 = 375 to 394 rpm with one engine inoperative

Rotor Speed Warning.

Aural at 360 rpm and 410 rpm.

Airspeed Limits.

Never exceed speed V_{NE} Power-on:
 150 kt zero pressure altitude. Doors open or removed: 70 knots at zero pressure altitude.

Never exceed speed V_{NE} Power-off
 120 kt at zero pressure-altitude
 See Rotorcraft Flight Manual for decrease of these values with altitude and temperature.

Minimum Crew.

1 pilot at 60.62 in. - Starboard seat

Maximum Passengers Standard (5).

4 at 98.42 in.
 1 at 60.62 in.
 Maximum passengers of 6 is available when the aircraft is equipped with the optional two-place seat at 60.62 in. This optional item is to be used in accordance with the associated Flight Manual supplement.

Maximum Baggage.

Right baggage compartment = 220 lb at 125.98 in
 Left baggage compartment = 264 lb at 125.908 in
 Rear baggage compartment = 176 lb at 181.10 in
 Main cabin (on rear = 682 lb at 88.58 in
 (on LH forward = 330 lb at 61.02 in

Fuel Capacity.

	Front Tank	Rear Tank
Total	88.1 U.S. gallons	106.5 U.S. gallons
Usable	87.2 U.S. gallons	105.7 U.S. gallons
C.G. Range	127.55 in	151.55 in

See NOTE for data on unusable fuel.

Empty Weight CG Range.

None

Leveling Means.

Transmission support platform

Oil Capacity.

Engine = 2 x 1.5 U.S. gallons at 140.5 in
 MGB = 2.9 U.S. gallons at 145.2 in
 TGB = 0.08 U.S. gallons at 379.5 in

Rotor Blade and Control Movements.

For rigging information, refer to the AS355 Maintenance Manual.

V. Model AS355N "Twinstar" (Normal Category), October 15, 1992.

The major difference between the AS355N and AS355F series is the replacement of the Allison C250 engines with Turbomeca Arrius 1 A engines.

Engine.

2 Turbomeca model Arrius 1 A controlled by a digital electronic control units.

Installed Engine Limits.

	Torque	Gas Generator Speed	Exhaust Gas Temp.	Output Shaft Speed
Take-off (5 min)	406 mN (78%)*	54,685 rpm	800°C	6,256 rpm
Maximum Continuous two engines operative	380 mN (73%)*	53,285 rpm	765°C	6,256 rpm
Maximum continuous one engine operative	521 mN (100%)*	53,285 rpm	765°C	6,256 rpm
30 mn. one engine inoperative	599 mN (115%)*	55,300 rpm	800°C	6,256 rpm
2 mn. 30s one engine inoperative	683 mN (131%)*	56,140 rpm	870°C	6,256 rpm

* Torque limits of helicopter main gearbox

Engine gearbox torque limit = 521 mN (100% torque)
 refer to Flight Manual AS355N for transients.

Fuel.

Refer to Flight Manual AS355N for approved fuels and additive specifications.

Helicopter Limits.

One engine inoperative:

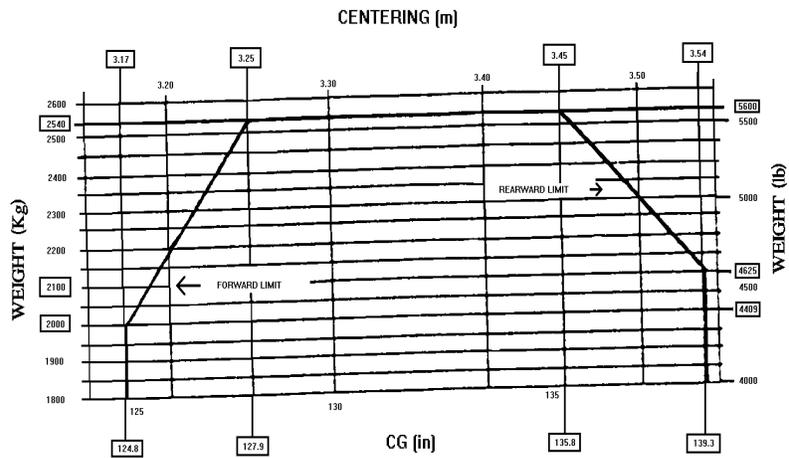
Maximum continuous torque: 100% (521 mN)
 30 mn. torque: 115% (599mN).
 2mn 30s. torque: 131% (683 mN).

Two engines operative:

IAS < 55 Kts.: 78% (406 mN).
 IAS > 55 Kts.: 73% (380 mN).

C.G. Range.

Longitudinal



Lateral - Right = 0.09 m (3.54 in)
 - Left = 0.16 m (6.30 in)

Datum.

Longitudinal = 3.40 m (133.8 in. forward of main rotor hub center)
 Lateral = Vertical plane passing longitudinally through main rotor hub center.

Maximum Weight.

5732 lb (2600 kg)

Rotor Speed Limitations.

Power-on flight
 - two engines in operation: 390 (+ 4, -5) rpm
 For IAS below 55 kt: 390 (+ 10, -5) rpm
 - OEI : 375 to 394 rpm

Power-off flight
 - max. 425 rpm (aural warning at 410 rpm)
 - min. 330 rpm (aural warning at 360 rpm)

Airspeed Limits.

Never exceed speed V_{NE} Power-on:
 150 kt zero pressure altitude.
 Doors removed: see the flight manual (section 2.1, § 5.2.)

Never exceed speed V_{NE} Power-off:
 120 kt at zero pressure-altitude.
 See Rotorcraft Flight Manual for decrease of these values with altitude and temperature.

Minimum Crew.

1 pilot at 60.62 in. - Starboard seat.

Maximum Passenger Standard (5).

4 at 98.42 in.
 1 at 60.62 in.
 Maximum passengers of 6 is available when the aircraft is equipped with the optional two-place seat at 60.62 in. This optional item is to be used in accordance with the associated Flight Manual supplement.

Maximum Baggage.

Right baggage compartment = 220 lb at 125.98 in
 Left baggage compartment = 264 lb at 125.908 in
 Rear baggage compartment = 176 lb at 181.10 in
 Main cabin (on rear = 682 lb at 88.58 in
 (on LH forward = 330 lb at 61.02 in

Fuel Capacity.

	Front Tank	Rear Tank
Total	88.1 U.S. gallons	106.5 U.S. gallons
Usable	87.2 U.S. gallons	105.7 U.S. gallons
C.G. Range	127.55 in	151.55 in

See NOTE 1 for data on unusable fuel.

Empty Weight CG Range.

None

Leveling Means.

Transmission support platform.

Oil Capacity.

Engine = 2 x 1.5 U.S. gallons at 140.5 in
 MGB = 2.9 U.S. gallons at 145.2 in
 TGN = 0.08 U.S. gallons at 379.5 in

Rotor Blade and Control Movements.

For rigging information, refer to the AS 355 Maintenance Manual.

VI. Model AS355NP "Twinstar" (Normal Category), January 9, 2008.

The major difference between the AS355NP and AS355N is the replacement of the Turbomeca Arrius 1A engines with Turbomeca Arrius 1A1 engines, addition of New engine monitoring system: VEMD (Vehicle and Engine Multifunction Display), electric oil cooling fan and upgraded main gear box.

Engine.

2 Turbomeca model Arrius 1A1 controlled by a digital electronic control units.

Installed Engine Limits.

	Torque	Gas Generator Speed	Exhaust Gas Temp.	Output Shaft Speed
Take-off (5 min)	450 mN (86.4%)*	54,375 rpm	773°C	6,016 rpm
Maximum Continuous two engines operative	374 mN (71.8%)*	53,285 rpm	749°C	6,016 rpm
Maximum continuous one engine operative	599 mN (115%)*	55,452 rpm	812°C	6,016 rpm
2 mn. 30s one engine inoperative	683 mN (131%)*	56,347 rpm	885°C	6,016 rpm

* Torque limits of helicopter main gearbox

Engine gearbox torque limit = 521 mN (100% torque)
 refer to Flight Manual AS355NP for transients.

Fuel.

Refer to Flight Manual AS355NP for approved fuels and additive specifications.

Helicopter Limits.

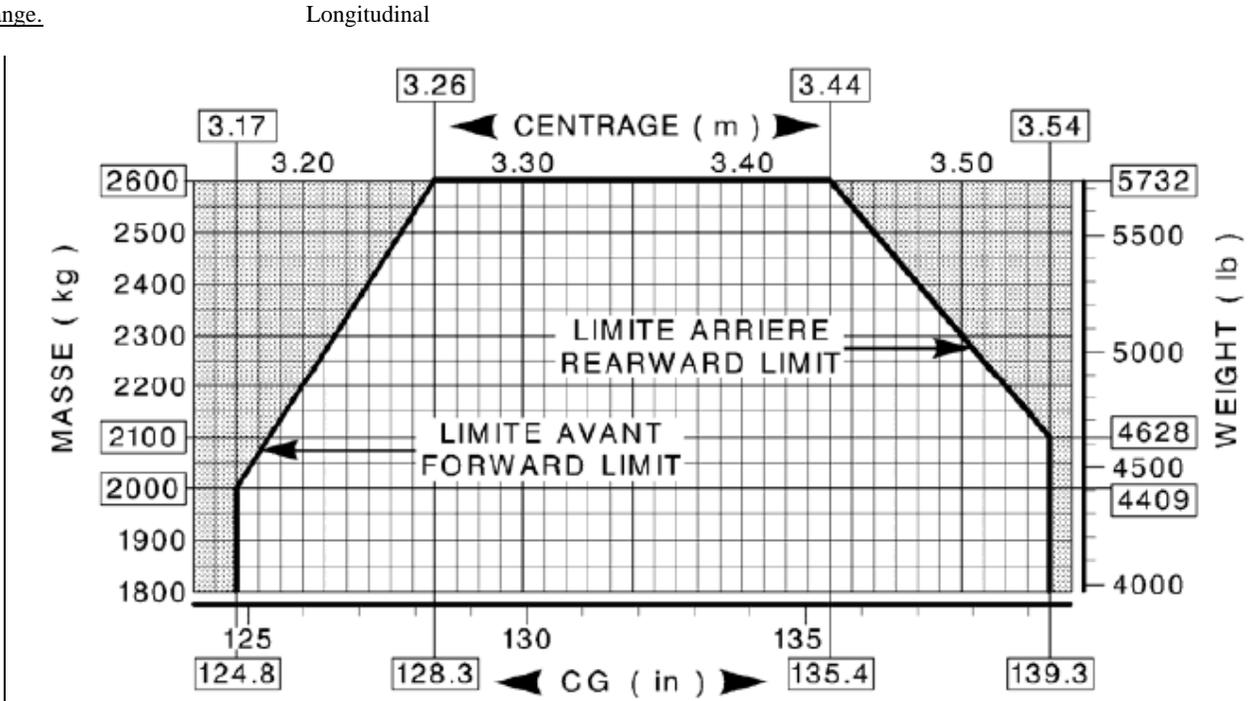
One engine inoperative:

Maximum continuous torque: 115% (599 mN)
 2mn 30s. torque: 131% (683 mN).

Two engines operative:

IAS < 55 Kts.: 86.4% (450 mN).
 IAS > 55 Kts.: 71.8% (374 mN).

C.G. Range.



- Lateral - Right = 0.09 m (3.54 in)
- Left = 0.09 m (3.54 in)
- Left = 0.16 m (6.30 in) (hoisting operation)

Datum.

Longitudinal = 3.40 m (133.8 in. forward of main rotor hub center)
 Lateral = Vertical plane passing longitudinally through main rotor hub center.

Maximum Weight.

5732 lb (2600 kg)

Rotor Speed Limitations.

Power-on flight
 - two engines in operation 375-394 rpm
 For IAS below 55 kt: 390 (+ 10, -5) rpm
 - OEI : 375 to 394 rpm

Power-off flight
 - max. 425 rpm (aural warning at 410 rpm)
 - min. 330 rpm (aural warning at 360 rpm)

Maximum Rotor Brake Application: 170 rpm

Airspeed Limits.

Never exceed speed V_{NE} Power-on:
 150 kt zero pressure altitude.
 Doors removed: see the flight manual (section 2, § 2.3)

Never exceed speed V_{NE} Power-off:
 120 kt at zero pressure-altitude.
 See Rotorcraft Flight Manual for decrease of these values with altitude and temperature.

Minimum Crew.

1 pilot at 60.62 in. - Starboard seat.

Maximum Passenger Standard(5).

4 at 98.42 in.
 1 at 60.62 in.

Maximum passengers of 6 is available when the aircraft is equipped with the optional two-place seat at 60.62 in. This optional item is to be used in accordance with the associated Flight Manual supplement.

Maximum Baggage.

Right baggage compartment = 220 lb at 125.98 in
 Left baggage compartment = 264 lb at 125.908 in
 Rear baggage compartment = 176 lb at 181.10 in
 Main cabin (on rear = 682 lb at 88.58 in
 (on LH forward = 330 lb at 61.02 in

Fuel Capacity.

	Front Tank	Rear Tank
Total	88.1 U.S. gallons	106.5 U.S. gallons
Usable	87.2 U.S. gallons	105.7 U.S. gallons
C.G. Range	127.55 in	151.55 in

See NOTE 1 for data on unusable fuel.

Empty Weight CG Range.

None

Leveling Means.

Transmission support platform.

Oil Capacity.

Engine = 2 x 1.5 U.S. gallons at 140.5 in
 MGB = 2.9 U.S. gallons at 145.2 in
 TGN = 0.08 U.S. gallons at 379.5 in

Rotor Blade and Control Movements.

For rigging information, refer to the AS 355 Maintenance Manual.

Certification Basis.

AS355E

AS355F, AS355F1, AS355F2

FAR 21.29 and FAR 27 effective February 1, 1965 plus Amendments 27-1 through 27-16.

Type Certificate No. H11EU, issued January 19, 1981. Amended November 20, 1981 to add Model AS355F. Amended April 11, 1984 to add Model AS355F1. Amended February 13, 1987 to add Model AS355F2. Date of Application for Type Certificate: January 4, 1979.

AS 355 N version:

- The basic certification basis consists of FAR 21.29 and FAR 27 effective February 1, 1965, including amendments 27.1 through 27.20, plus
- The following sections of amendment 27.21:
 27.21, 27.45, 27.71, 27.79, 27.143, 27.151, 27.161, 27.173, 27.175, 27.177, 27.672, 27.673, 27.729, 27.735, 27.779, 27.807, 27.1329, 27.1413, 27.1519, 27.1525, 27.1555, 27.1585 and 27.1587.

DGAC special conditions notified in letter DGAC No. 54408 dated October 21, 1988 are:

- A - Limit Pilot Forces
- B - Engine air intake protection against birds (2 lbs) and hail ingestion
- C - Engine governing system

The FADEC has been HIRF tested on the Arrius 1A engine with success at 200V/m according to Cat Y DO160.

In this certification basis, the following sections remain at the Amendment preceding the Amendment 27.21: 27.141, 27.610, 27.785, 27.1309, 27.1505 and 27.1559.

The French Direction Generale de l'Aviation Civile (DGAC) originally type certificated this rotorcraft under its type certificate TC 168. The FAA validated this product under U.S. Type Certificate Number H11EU. Effective September 28, 2003, the European Aviation Safety Agency (EASA) began oversight of this product on behalf of the DGAC.

AS 355 NP version:

- The basic certification basis consists of FAR 21.29 and FAR 27 effective February 1, 1965, including amendments 27.1 through 27.20, plus
- The following sections of amendment 27.21:
27.21, 27.45, 27.71, 27.79, 27.143, 27.151, 27.161, 27.173, 27.175, 27.177, 27.672, 27.673, 27.729, 27.735, 27.779, 27.807, 27.1329, 27.1413, 27.1519, 27.1525, 27.1555, 27.1585 and 27.1587.
- The following section of amendment 27.23: 27.923

DGAC special conditions notified in letter DGAC No. 54408 dated October 21, 1988 are:

- A - Limit Pilot Forces
- B - Engine air intake protection against birds (2 lbs) and hail ingestion
- C - Engine governing system

The FADEC HIRF approval for the Arrius 1A1 was accomplished by similarity to the testing accomplished on the Arrius 1A engine installed on the AS355N helicopter.

The AS355NP also complies with the following FAR 29 paragraphs: §45(a) and (b)(2) Amdnt 24; §49(a) Amdnt 39; §51 Amdnt 39; §53 Amdnt 39; §55 Amdnt 39; §59 Amdnt 44; §60 Amdnt 39; §61 Amdnt 39; §62 Amdnt 44; §64 Amdnt 39; §65(a) Amdnt 39; §67(a) Amdnt 44; §75 Amdnt 39; §77 Amdnt 44; §79 Amdnt 39; §81 Amdnt 44; §85 Amdnt 44; §87(a) Amdnt 39; §861(a) Amdnt 30; §901(c) Amdnt 26; §903(b) (c) and (e) Amdnt 36; §908(a) Amdnt 26; §917(c)(1) Rotor drive system: Design. Amdnt 40; §953(a) Amdnt 0; §1027(a) Amdnt 26; §1045(a)(1), (b), (c), (d), and (f) Amdnt 26; §1047(a) Amdnt 26; §1181(a) Amdnt 26; §1187(e) Amdnt 0; §1189(c) Amdnt 26; §1191(a)(1) Amdnt 3; §1193(e) Amdnt 26; §1195(a) and (d) Amdnt 17; §1197 Amdnt 13; §1199 Amdnt 13; §1201 Amdnt 0; §1305 (b) Amdnt 40; §1309(b) Amdnt 14; §1323(c)(1) Amdnt 44; §1331(b) Amdnt 24; §1587(a) Amdnt 44.

FAR Part 36 Noise Standards amended by Amendments 36-1 through 36-21.

DATA PERTINENT TO ALL MODELS.

Serial Numbers Eligible.

The French Government "Certificat de Navigabilite pour Exportation" endorsed as noted below under "Import Requirements" must be submitted for each individual aircraft for which application for FAA certification is made.

Type Certificate.

Type Certificate No. H11EU, issued January 19, 1981.
Amended November 20, 1981 to add Model AS355F.
Amended April 11, 1984 to add Model AS355F1.
Amended February 13, 1987 to add Model AS355F2.
Amended October 15, 1992 to add Model AS355N. Date of application for amended type certificate: February 12, 1991.
Amended January 9, 2008 to add Model AS355NP. Date of application for amended type certificate: April 6, 2007.

Import Requirements.

The FAA can issue a U.S. airworthiness certificate based on a National Aviation Authority (NAA) Export Certificate of Airworthiness (Export C of A) signed by

a representative of the French Generale de l'Aviation Civile (DGAC) on behalf of the European Community.

The Export C of A should contain the following statement: "The aircraft covered by this certificate has been examined, tested, and found to comply with the type design approved under U.S. Type Certificate Number H11EU and to be in a condition for safe operation."

Service Information.

Each of the documents listed below must state that it is approved by the European Aviation Safety Agency (EASA) or – for approvals made before September 28, 2003 – by the French Generale de l'Aviation Civile (DGAC). Any such documents are accepted by the FAA and are considered FAA approved.

- Service Bulletin,
- Structural repair manuals,
- Vendor manuals,
- Aircraft flight manuals, and
- Overhaul and maintenance manuals.

This applies only to the acceptance of the type design data."

Equipment.

The basic required equipment as prescribed in the applicable airworthiness regulations (See Certification Basis) must be installed in the helicopter for certification. Eurocopter Report No. 355A043462 lists required and optional equipment for the helicopter.

In addition, the following item of equipment is required.

EASA approved Rotorcraft Flight Manual, identified as Code B:

1. For AS355E, Issue 1, Amendment 0, approved January 16, 1981 or later approved revision.
2. For AS355F, Revision 0 approved April 14, 1981.
Coded 81.16 or later approved revision.
3. For AS355F1, Revision 0 approved March 15, 1984.
Coded 84.11 or later approved revision.
4. For AS355F2, Revision 0 approved February 13, 1987 plus Rush Revision 1B and specific pages marked B or later approved revision.
5. For AS355N, Revision 2 approved October 2, 1992 and specific pages marked B or later approved revision.
6. For AS355NP, Revision 0 approved January 9, 2008, with FAA unique pages marked B, or later approved revision.

NOTES

NOTE 1.

Current weight and balance report including loading instructions and list of equipment included in the certificated empty weight, must be provided for each helicopter at the time of original certification. The certificated empty weight and corresponding center of gravity location must include unusable fuel of 6.1 lb at 127.55 in for Front tank and 5.6 lb at 151.55 in for Rear tank.

In order to obtain the most consistent weight and balance results, all helicopters should be weighed on jackpoints rather than on wheels and floats. When changes are made to the helicopter which affect the weight and balance, refer to the Flight Manual Weight and Balance Appendix for

instructions.

NOTE 2. All placards indicated in the Rotorcraft Flight Manual must be installed in the appropriate location.

NOTE 3. Information essential to the proper maintenance of the helicopter is contained in the manufacturer's AS355 Maintenance Manual provided with each helicopter. Life-limited components and associated retirement times are presented in Chapter 5, Section CD 5.99, and must be replaced in accordance therewith.

NOTE 4. Service Bulletin No. 01.02 to be used to convert 355E version of 355F version (VFR).

NOTE 5. Service Bulletin No. 01.09 to be used to convert 355F version to 355F1 version.

NOTE 6. Model AS355F2 must be equipped with an approved autorelight system to be eligible for IFR certification below + 5° C. OAT.

.....END.....