

[Federal Register Volume 79, Number 192 (Friday, October 3, 2014)]  
[Rules and Regulations]  
[Pages 59643-59645]  
From the Federal Register Online via the Government Printing Office [www.gpo.gov]  
[FR Doc No: 2014-23378]

---

## **DEPARTMENT OF TRANSPORTATION**

### **Federal Aviation Administration**

#### **14 CFR Part 39**

**[Docket No. FAA-2014-0290; Directorate Identifier 2012-NM-210-AD; Amendment 39-17981; AD 2014-20-08]**

**RIN 2120-AA64**

### **Airworthiness Directives; Lockheed Martin Corporation/Lockheed Martin Aeronautics Company Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Final rule.

---

**SUMMARY:** We are adopting a new airworthiness directive (AD) for certain Lockheed Martin Corporation/Lockheed Martin Aeronautics Company Model L-1011 series airplanes. This AD was prompted by reports of cracked rib cap castellations. This AD requires repetitive inspections for castellation and skin clips cracked or damaged between stringers and cracked stringer clips of the wing box pylon back-up structure, and front spar to rear spar; repetitive inspections for cracking, damage, or failure of the pylon back-up torque box structure; repetitive inspections for cracking or damage of the wing box external areas at the drag brace aft wing fitting; repetitive inspections of the outer surface of the wing upper and lower skins for cracks or damage along the rib attachment at the fastener holes and between the two rows of attachment; and corrective actions if necessary. We are issuing this AD to detect and correct cracked or damaged rib cap castellations, which could degrade the structural capabilities of the airplane.

**DATES:** This AD is effective November 7, 2014.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of November 7, 2014.

**ADDRESSES:** For service information identified in this AD, contact Lockheed Martin Corporation/Lockheed Martin Aeronautics Company, L1011 Technical Support Center, Dept. 6A4M, Zone 0579, 86 South Cobb Drive, Marietta, GA 30063-0579; telephone 770-494-5444; fax 770-494-5445; email L1011.support@lmco.com; Internet <http://www.lockheedmartin.com/ams/tools/TechPubs.html>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

## **Examining the AD Docket**

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2014-0290; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800-647-5527) is Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

**FOR FURTHER INFORMATION CONTACT:** Carl Gray, Aerospace Engineer, Airframe Branch, ACE-117A, FAA, Atlanta Aircraft Certification Office (ACO), 1701 Columbia Avenue, College Park, GA 30337; phone: 404-474-5554; fax: 404-474-5605; email: [carl.w.gray@faa.gov](mailto:carl.w.gray@faa.gov).

## **SUPPLEMENTARY INFORMATION:**

### **Discussion**

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 by adding an AD that would apply to certain Lockheed Martin Corporation/Lockheed Martin Aeronautics Company Model L-1011 series airplanes. The NPRM published in the Federal Register on May 29, 2014 (79 FR 30748). The NPRM was prompted by reports of cracked rib cap castellations. The NPRM proposed to require repetitive inspections for castellation and skin clips cracked or damaged between stringers and cracked stringer clips of the wing box pylon back-up structure, and front spar to rear spar; repetitive inspections for cracking, damage, or failure of the pylon back-up torque box structure; repetitive inspections for cracking or damage of the wing box external areas at the drag brace aft wing fitting; repetitive inspections of the outer surface of the wing upper and lower skins for cracks or damage along the rib attachment at the fastener holes and between the two rows of attachment; and corrective actions if necessary. We are issuing this AD to detect and correct cracked or damaged rib cap castellations, which could degrade the structural capabilities of the airplane.

### **Comments**

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM (79 FR 30748, May 29, 2014) or on the determination of the cost to the public.

### **Conclusion**

We reviewed the relevant data and determined that air safety and the public interest require adopting this AD as proposed except for minor editorial changes. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM (79 FR 30748, May 29, 2014) for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM (79 FR 30748, May 29, 2014).

### **Costs of Compliance**

We estimate that this AD affects 26 airplanes of U.S. registry.  
We estimate the following costs to comply with this AD:

### Estimated Costs

Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
Inspections	41 work-hours × \$85 per hour = \$3,485 per inspection cycle	\$0	\$3,485 per inspection cycle	\$90,610 per inspection cycle.

We estimate the following costs to do any necessary repairs that would be required based on the results of the inspections. We have no way of determining the number of aircraft that might need these repairs:

### On-Condition Costs

Action	Labor cost	Parts cost	Cost per product
Modification (up to 12 rib caps per airplane)	96 work-hours × \$85 per hour = \$8,160 per rib cap	\$15,000 per rib cap	\$23,160 per rib cap.

Other than the modification stated above, we have received no definitive data that would enable us to provide cost estimates for the crack repair actions specified in this AD.

### Paperwork Reduction Act

A federal agency may not conduct or sponsor, and a person is not required to respond to, nor shall a person be subject to penalty for failure to comply with a collection of information subject to the requirements of the Paperwork Reduction Act unless that collection of information displays a current valid OMB control number. The control number for the collection of information required by this AD is 2120-0056. The paperwork cost associated with this AD has been detailed in the Costs of Compliance section of this document and includes time for reviewing instructions, as well as completing and reviewing the collection of information. Therefore, all reporting associated with this AD is mandatory. Comments concerning the accuracy of this burden and suggestions for reducing the burden should be directed to the FAA at 800 Independence Ave. SW., Washington, DC 20591, ATTN: Information Collection Clearance Officer, AES-200.

### Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

### Regulatory Findings

This AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and

the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866,
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

### **List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

### **Adoption of the Amendment**

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

### **PART 39—AIRWORTHINESS DIRECTIVES**

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

#### **§ 39.13 [Amended]**

2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):



---

**2014-20-08 Lockheed Martin Corporation/Lockheed Martin Aeronautics Company:**  
Amendment 39-17981; Docket No. FAA-2014-0290; Directorate Identifier 2012-NM-210-AD.

**(a) Effective Date**

This AD is effective November 7, 2014.

**(b) Affected ADs**

None.

**(c) Applicability**

This AD applies to Lockheed Martin Corporation/Lockheed Martin Aeronautics Company Model L-1011-385-1, L-1011-385-1-14, L-1011-385-1-15, and L-1011-385-3 airplanes, certificated in any category, as identified in Lockheed Martin Service Bulletin 093-57-207, Revision 5, dated November 14, 2008.

**(d) Subject**

Air Transport Association (ATA) of America Code 57, Wings.

**(e) Unsafe Condition**

This AD was prompted by reports of cracked rib cap castellations. We are issuing this AD to detect and correct cracked or damaged rib cap castellations, which could degrade the structural capabilities of the airplane.

**(f) Compliance**

Comply with this AD within the compliance times specified, unless already done.

**(g) Repetitive Wing Inspections**

For Model L-1011-385-1, L-1011-385-1-14, L-1011-385-1-15, and L-1011-385-3 airplanes, serial numbers 1189 and subsequent: At the applicable compliance time specified in paragraphs (h)(1), (h)(2), and (h)(3) of this AD, do the inspections specified in paragraphs (g)(1) through (g)(4) of this AD. Repeat the inspections thereafter at intervals not to exceed 3,600 flight cycles or 7,200 flight hours, whichever occurs first.

(1) Do a detailed inspection for castellation and skin clips cracked or damaged (including cracks, loose or missing fasteners, oversized and missed drilled fastener holes, corrosion, dents, scratches and other signs of distress) between stringers and cracked stringer clips of the wing box pylon back-up structure, and front spar to rear spar, in accordance with the Accomplishment Instructions of Lockheed Martin Service Bulletin 093-57-207, Revision 5, dated November 14, 2008.

(2) Do a general visual inspection for cracking or damage (including cracks, loose or missing fasteners, oversized and missed drilled fastener holes, corrosion, dents, scratches and other signs of distress) of the pylon back-up torque box structure, in accordance with the Accomplishment Instructions of Lockheed Martin Service Bulletin 093-57-207, Revision 5, dated November 14, 2008.

(3) Do a general visual inspection for cracking, damage (including cracks, loose or missing fasteners, oversized and missed drilled fastener holes, corrosion, dents, scratches and other signs of distress), or failure of the wing box external areas at the drag brace aft wing fitting, in accordance with the Accomplishment Instructions of Lockheed Martin Service Bulletin 093-57-207, Revision 5, dated November 14, 2008.

(4) Do a general visual inspection for cracking or damage (including cracks, loose or missing fasteners, oversized and missed drilled fastener holes, corrosion, dents, scratches and other signs of distress) of the outer surface of the wing upper and lower skins for cracks along the rib attachment at the fastener holes and between the two rows of attachments, in accordance with the Accomplishment Instructions of Lockheed Martin Service Bulletin 093-57-207, Revision 5, dated November 14, 2008.

#### **(h) Compliance Times for the Actions Specified in Paragraph (g) of This AD**

(1) For airplanes that have not accomplished the inspections described in Lockheed Martin Service Bulletin 093-57-207 prior to the effective date of this AD: At the later of the compliance times specified in paragraphs (h)(1)(i) and (h)(1)(ii) of this AD.

(i) Before the accumulation of 15,000 total flight cycles or 27,000 total flight hours, whichever occurs first.

(ii) Within 1,800 flight cycles or 3,600 flight hours, whichever occurs first, after the effective date of this AD.

(2) For airplanes that have accomplished the inspections described in Lockheed Martin Service Bulletin 093-57-207 prior to the effective date of this AD: Within 3,600 flight cycles or 7,200 flight hours, whichever occurs first, after the completion of the most recent inspections, except as specified in paragraph (h)(3) of this AD.

(3) For rib caps that have been modified as described in Lockheed Martin Service Bulletin 093-57-207: Before the accumulation of 15,000 total flight cycles or 27,000 total flight hours, whichever occurs first, for that rib cap only.

#### **(i) Corrective Actions**

If any cracking, damage, or failure is found during any inspection required by paragraph (g) of this AD: Before further flight, do all applicable corrective actions, in accordance with the Accomplishment Instructions of Lockheed Martin Service Bulletin 093-57-207, Revision 5, dated November 14, 2008, except where this service bulletin specifies that all other damaged structural items should be repaired using the best shop practices, following procedures in Structural Repair Manual 57-12-00, this AD requires repairing the damage before further flight, in accordance with a method approved by the Manager, Atlanta Aircraft Certification Office (ACO), FAA. For a repair method to be approved by the Manager, Atlanta ACO, as required by this paragraph, the Manager's approval letter must specifically refer to this AD.

#### **(j) Reporting**

Submit a report of positive findings of the inspection for cracking, damage, or failure required by this AD to the Manager, Atlanta ACO, at the applicable time specified in paragraph (j)(1) or (j)(2) of this AD. The report must include the inspection results, a description of the discrepancies found, the airplane serial number, and the number of landings and flight hours on the airplane.

(1) If the inspection was done on or after the effective date of this AD: Submit the report within 30 days after the inspection.

(2) If the inspection was done before the effective date of this AD: Submit the report within 30 days after the effective date of this AD.

**(k) Paperwork Reduction Act Burden Statement**

A federal agency may not conduct or sponsor, and a person is not required to respond to, nor shall a person be subject to a penalty for failure to comply with a collection of information subject to the requirements of the Paperwork Reduction Act unless that collection of information displays a current valid OMB Control Number. The OMB Control Number for this information collection is 2120-0056. Public reporting for this collection of information is estimated to be approximately 5 minutes per response, including the time for reviewing instructions, completing and reviewing the collection of information. All responses to this collection of information are mandatory. Comments concerning the accuracy of this burden and suggestions for reducing the burden should be directed to the FAA at: 800 Independence Ave. SW., Washington, DC 20591, Attn: Information Collection Clearance Officer, AES-200.

**(l) Credit for Previous Actions**

This paragraph provides credit for actions required by paragraphs (g) and (i) of this AD, if those actions were performed before the effective date of this AD using Lockheed Martin Service Bulletin 093-57-207, Revision 3, dated November 22, 1991.

**(m) Alternative Methods of Compliance (AMOCs)**

(1) The Manager, Atlanta ACO, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ACO, send it to the attention of the person identified in paragraph (n)(1) of this AD.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

**(n) Related Information**

(1) For more information about this AD, contact Carl Gray, Aerospace Engineer, Airframe Branch, ACE-117A, FAA, Atlanta Aircraft Certification Office (ACO), 1701 Columbia Avenue, College Park, GA 30337; phone: 404-474-5554; fax: 404-474-5605; email: carl.w.gray@faa.gov.

(2) Service information identified in this AD that is not incorporated by reference is available at the addresses specified in paragraphs (o)(3) and (o)(4) of this AD.

**(o) Material Incorporated by Reference**

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) Lockheed Martin Service Bulletin 093-57-207, Revision 5, dated November 14, 2008.

(ii) Reserved.

(3) For Lockheed service information identified in this AD, contact Lockheed Martin Corporation/Lockheed Martin Aeronautics Company, L1011 Technical Support Center, Dept. 6A4M, Zone 0579, 86 South Cobb Drive, Marietta, GA 30063-0579; telephone 770-494-5444; fax 770-494-

5445; email L1011.support@lmco.com; Internet  
<http://www.lockheedmartin.com/ams/tools/TechPubs.html>.

(4) You may view this service information at FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, Washington. For information on the availability of this material at the FAA, call 425-227-1221.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on September 23, 2014.

Dionne Palermo,  
Acting Manager, Transport Airplane Directorate,  
Aircraft Certification Service.