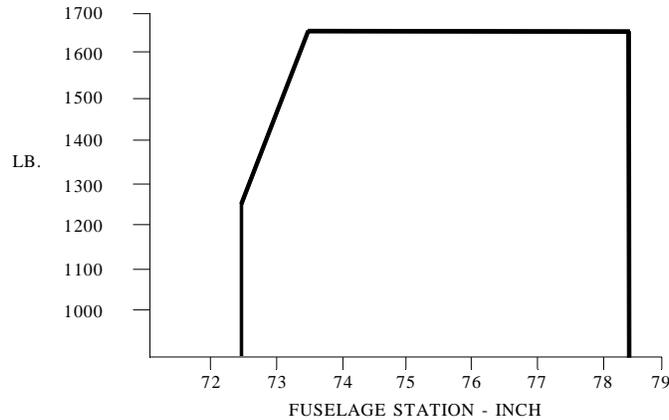




Center of Gravity RangeNormal and Utility Category

(+72.4) to (+78.5) at 1277 lb. or less  
 (+73.5) to (+78.5) at 1670 lb.

Straight line variation between points given.

Empty Weight C. G. Range

None

Datum

66.25 inches forward of wing leading edge.

Leveling Means

Top of tailcone behind rear window.

Maximum Weight

1670 lb. (Normal and Utility Category)

Number of Seats

2 (2 at +85.5)

Maximum Baggage

100 lb. (+115.0)

Fuel Capacity

32 gallons at (+75.4) (2 wing tanks)  
 See NOTE 1 for data on unusable fuel.

Oil Capacity

6 quarts at (+24.4) (4 quarts usable)  
 See NOTE 1 for data on unusable oil.

Control Surface Movements

Wing Flaps	(± 2°)	Up	0°	Down	34°
Ailerons	(± 2°)	Up	26°	Down	14°
Rudder	(± 1°)	Left	29°	Right	29°
Elevator	(± 2°)	Up	34°	Down	20°

Nose Wheel Travel

(± 2°)	Left	30°	Right	30°
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Serial Numbers Eligible

38-78A0001 through 38-82A0122 (See NOTE 6).

Certification Basis

Type Certificate Number A18SO, issued December 20, 1977, obtained by the manufacturer under delegation option procedures.

Date of application for Type Certificate, June 1, 1976.

FAR Part 23, effective February 1, 1965, as amended by Amendment by 23-16, effective February 14, 1975, and FAR 36 including Amendments through 36-7, effective October 1, 1977.

Finding of equivalent safety was made for the airspeed indicator (FAR 23.1545(a)).

Production Basis

Approved for manufacture of spare parts only under Production Certificate No. 206.

Equipment

The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

In addition, the following document is required:

1. Pilot's Operating Handbook, Piper Report Number 2126, dated January 20, 1978.

NOTE 1. Current weight and balance report, including list of equipment included in certificated empty weight, and loading instructions when necessary, must be provided for each aircraft at the time of original certification.

The certificated empty weight and corresponding center-of-gravity locations must include undrainable system oil (not included in oil capacity) and unusable fuel as noted below:

Unusable Fuel & Oil Quantity

Fuel 12.0 lb. at (+75.4)

Oil 1.90 lb. at (+24.4)

NOTE 2. The following placard must be displayed in clear view of the pilot:

"EXCEPT AS MAY BE OTHERWISE INDICATED ON A PLACARD, THE MARKINGS AND PLACARDS INSTALLED IN THIS AIRPLANE CONTAIN OPERATING LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THIS AIRPLANE IN THE NORMAL CATEGORY. OTHER OPERATING LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THIS AIRPLANE IN THIS CATEGORY OR IN THE UTILITY CATEGORY ARE CONTAINED IN THE PILOT'S OPERATING HANDBOOK.

FLAPS UP SPINS ARE APPROVED FOR UTILITY CATEGORY OPERATION. FOR SPIN RECOVERY, USE FULL RUDDER AGAINST THE SPIN FOLLOWED IMMEDIATELY BY FORWARD WHEEL.

THIS AIRPLANE IS APPROVED FOR FLIGHT IN DAY/NIGHT VFR/IFR WHEN EQUIPPED IN ACCORDANCE WITH F.A.R. 91. FLIGHT INTO KNOWN ICING CONDITIONS IS PROHIBITED.

FLAPS EXTENDED SPINS PROHIBITED."

NOTE 3.

PA-38-112 - all S/N:

- (a.) The service life of the wing and associated structure has been established as 11,000 hours time-in-service (TIS).
- (b.) The lower longitudinal trim springs, Piper P/N 37523 or 61916-2, have a life limit of 1,500 hours TIS.
- (c.) The service life of the steel upper rudder hinge, Piper P/N 77610-03, has been established as 5,000 hours TIS.

PA-38-112 - S/N 38-78A0001 through 38-80A0198:

The service life of the forward fin spar attachment plate, Piper P/N 77553-05, has been established as 3,000 hours TIS.

PA-38-112 - S/N 38-81A0001 through 38-82A0101:

- (a.) Aircraft not equipped with Piper Kit No. 764-421, the service life of the forward fin spar attachment plate, Piper P/N 77553-05, has been established as 3,000 hours TIS.
- (b.) Aircraft equipped with Piper Kit No. 764-421, the service life of the forward fin spar attachment plate, Piper P/N 77553-05, has been established as 5,000 hours TIS.

PA-38-112 - S/N 38-82A0102 through 38-82A0122:

The service life of the forward fin spar attachment plate, Piper P/N 77553-05, has been established as 5,000 hours TIS.

NOTE 4.

The Model PA-38-112, S/N 38-78A0001 through 38-82A0122 may be operated with the spinner dome removed, or with the spinner dome and rear bulkhead removed.

NOTE 5.

Inboard flow strips installed per Piper Drawing No. 61927-2, are installed on S/N 38-78A0017, 38-78A0786, 38-78A0812, and 38-78A0815. All S/Ns are eligible for installation of inboard flow strips per Piper Drawing No. 61927-2.

Pilot's Operating Handbook, Piper Report Number 2126, Revision 3, is required with this installation.

NOTE 6. The following serial numbers are not eligible for airworthiness certification in the United States:

38-78A0344, 38-78A0358, 38-78A0424, 38-78A0439, 38-78A0624, 38-78A0639, 38-78A0734, 38-78A0749, 38-78A0829, 38-78A0844, 38-79A0058, 38-79A0088, 38-79A0156, 38-79A0221, 38-79A0236, 38-79A0271, 38-79A0301, 38-79A0699, 38-79A0713, 38-79A1042, , 38-79A1051, 38-79A1056, 38-79A1061, 38-79A1115, 38-79A1117, 38-79A1122, 38-79A1126, 38-79A1131, 38-79A1134, 38-79A1158, 38-79A1162, 38-79A1165, 38-79A1168, 38-80A0015, 38-80A0019, 38-80A0022, 38-80A0025, 38-80A0035, 38-80A0038 through 38-80A0051, 38-80A0054, 38-80A0056, 38-80A0057, 38-80A0060, 38-80A0063, 38-80A0077, 38-80A0080, 38-80A0083, 38-80A0086, 38-80A0104, 38-80A0113, 38-80A0120, 38-80A0123 through 38-80A0165, 38-80A0169 through 38-80A0198, 38-81A0136 through 38-81A0138, and 38-81A0191 through 38-81A0210.

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